

SPIS simulations in support of science projects and their limitations

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Introduction



- SPIS has been called upon to aid the design of scientific spacecraft
 - To quantify the disturbance to the environment caused by the spacecraft
 - To assess contamination of measurements from the presence of the spacecraft and active sources
 - To optimise the location of instruments
 - To aid in the interpretation of observations.
- Despite success, there have been difficulties in using SPIS for this purpose
- Some examples from Swarm, Cluster and MMS are given
- Probe simulations have shown unexpected differences to theory

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- SWARM a multi-satellite mission to survey Earth's geomagnetic field and its temporal evolution
- 3 satellites in three different polar orbits between 400 and 550 km altitude.
- High-precision and high-resolution measurements of the strength and direction of the magnetic field will be made
- Electric fields can influence measurements taken by the sensitive magnetometers on board.
- Goal of SPIS simulation: to help quantify the disturbance that would be experienced by the Langmuir probe on the SWARM spacecraft

SWARM





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- We could not directly simulate the behaviour of the Langmuir probe
- Sources of interference of Langmuir probe
 - a. Geometrical blocking due to nearby spacecraft
 - b. Electric field due to presence of spacecraft
- Point a was not addressed
- Point **b** was addressed
 - Depends on s/c surface potential and geometry
 - Depends on mainly plasma characteristics and the current collection on conductive elements
 - Metallic plate on ram (of size to be determined)
 - Metallic interconnects of solar array
- SPIS could not simulate the thousands of tiny solar array interconnects, so the same area was used in larger patches











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n=9E4cm-3, Te=0.237, Ti=0.144, Debye length 12mm Vel=7.7km/s, Species O+

Floating potential: -0.50V with a 0.35m² ram plate -0.49V with no ram plate

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SWARM



How accurate are these SPIS results given the interconnect approximation?

Interconnects were mostly 2mmx2mm in size

Equilibrium is negative, and so isotropic electrons are well simulated by Boltzmann equation



Combining small interconnects into larger surfaces moves ion current collection from the thick sheath case towards thin sheath.

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Ion Current-Voltage curve lies between thin and thick sheath approximations

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- Comparison of SPIS with theoretical calculations

Ram plate cm ²	SPIS V	Thin V	Thick/10 V
0.35	-0.5	-0.55	-0.51
0	-0.49	-0.48	-0.18

- Better to include I-V curves of solar arrays in SPIS

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- Cluster wake crossings contaminate electric field measurements
- A pulse of duration 0.3s and amplitude 1.5V occurs twice per rotation

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Electron Temp (eV)	8.61
lon Temp. (eV)	10.34
Ion/Electron density (m ⁻³)	8.7x10 ⁶
Drift (km/s)	468



The full wake region could not be simulated in high resolution

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- E-field is calculated as V/88m
- Pulse of 0.3s duration and about 1.5V/m is predicted
- However, a background potential was produced which may be artefact of coarse wake simulation and narrow simulation volume







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MMS





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Spherical probe simulation



Validation test case

Sphere in large Debye length regime:A systematic difference of 5% found.No explanation.

Quantity	Symbol	Value	
Temperature	Т	1.0	eV
Electron density	n	10^{7}	m^{-3}
Debye length	λ	~2.35	m
Potential	φ	[0,12]	V
Sphere radius	ρ	0.25	m
Number of tetrahedrons		~30,000	
Simulation box diameter		20	m
Number of macro-particles		~300,000	

Ref.: Hilgers et al., IEEE Trans. Plasma Sc., 36, 5, 2008.



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Spherical probe simulation



Backtracking (SPIS v4.3)

12000 particles tracked

Sphere in large Debyelength regime:At 0V virtually exactAt higher voltages upto 2.5% error



Cylindrical probe in Large Debye length regime



Cylinder in large Debye length regime:

- •Very large difference is found.
- •Finite length effect
- •Is trajectory calculation accurate enough?



Cylindrical probe simulation





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Conclusion



- SPIS has allowed the effects of the spacecraft presence on measurements to be investigated.
- Not all the details that are wanted could be simulated (e.g. interconnects and high resolution everywhere)
- In particular the induced potential due to spacecraft floating potential at the instrument location can be assessed
- There is uncertainty in the neutralisation of ion plumes
- We want to simulate what the instrument itself measures, e.g. current for Langmuir probes, particle distributions for electrostatic analysers.
- By developing the reverse trajectory analysis capabilities of SPIS, this may be possible
- BUT we need to understand differences with probe theory

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Thank you for your attention





Cylindrical probe with +8V

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Solar probe





 Spacecraft is in shadow and sensors on booms extend beyond a sun-shield.

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Solar probe





- Smaller simulation space than for Cluster allows better statistics
- Forcing shadowed surfaces to have no photoemission was difficult

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Solar probe







 Instruments on short booms are out of the wake but influenced by the spacecraft potential

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